

* *
2007/10/22 :
2008/4/3 :

. (10.7mm) (12.7mm)
 (() (1m)
 .(20 kW/m²,10 kW/m²)
 ((20()21 ())
 (Ansys.Ver.9) (F.E.M)

Abstract

In this work theoretical study is carried out to the effect of heat flux on induced vibrations from a fluid flow in a simply and flexible supported pipe with a restriction. The water flow in the pipe is assumed to be fully developed region. The dynamics of a pipe conveying fluid is described by the transfer matrix approach. It provides a numerical technique for solving the equations of motion of forced and free vibrations for simply and flexible support. The pipe is divided into (21) nodes and (20) element along its length. Then, from the pipe analysis equation, a fortran Language computer program has been built to embrace the theoretical and analytical work. And then to ensure that the program developing is suitable, it has been compared with another program called (ANSYS Ver.9) which it use FEM. The Obtained results show that the transfer matrix method is a suitable technique to compute the eigen modes in addition to the natural frequency of vibration systems. From this research it has been found that the heat flux affects the natural frequencies and the amplitude of vibration. So the increasing the thermal force results in the decreasing of the magnitude of the natural frequencies. The amplitudes of vibration along the pipe increase as the excitation frequency is so close to the natural frequencies of the system, and this effect is predominate than the effect of the heat flux.

Keyword: (Effect of Heat Flux On Induced Vibrations, Fluid Flow In A pipe With A Restriction At Mid The Pipe)

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m^2		A_f
m^2		A_p
—		A
m^2	(Vena Contracta)	A_2
m		a
m		D
N/m^2		E
—		$[\bar{F}]_i$
N		F_i
N/m^2		G
$W/m^2 \cdot c^\circ$		h
m^4		I
$W/m \cdot c^\circ$		k_F
kg/m		m_f
kg/m		m_p
N		N
—		N_{Ud}
N/m^2		P
—		$[\bar{P}]_i$
W/m^2		q
C°		T_w
C°	(Bulk . Temperature)	T_b
m/s		V_f
m/s		V_J
N	+	W

—		$[\bar{Z}]_i$

rad/s		ω_n
$1/C^\circ$		α_p

	f
	J
	i
	p

	R , L
	—

[1] ()

(Oil pipe line)

(Little and

.Weaver)[2]

(Rockets and Aircraft)

(Heat Exchanger)

.(Roberrt and Chung)[3]

(Heat Exchanger)

(Jet Engine)

(Resonance Peaks)

(2001)

(Alaa.A.M.H) [5]

:

-1

-2

-3

:

(Compressive Force)

-a

(Rotational Force)

-b

(2003)

(Yang and Soon) [6]

)

(

.(Transfer Matrix Method)

:

-1

-2

(Re_v)(Vibrational Reynolds Number)

$$Re_v = \frac{a \cdot \omega \cdot D}{\nu}$$

-3

(1989)

(Shigea ,atel) [4]

(2005)

(Zena) [7]

(double Dynamic absorber)

$$EI \frac{\partial^4 Y}{\partial X^4} + N \frac{\partial^2 Y}{\partial X^2} + [m_f V_f V_j + P A_f] \frac{\partial^2 Y}{\partial X^2} + 2m_f V_f \frac{\partial^2 Y}{\partial X \partial t} + (m_f + m_p) \frac{\partial^2 Y}{\partial t^2} = F(x, t) \quad \dots\dots\dots (1)$$

F(x, t)

(2006)

: [10] (δT)

(Ansam) [8]

$$\delta T = \alpha_p \cdot L \cdot \Delta T \quad \dots\dots\dots (2)$$

$$\delta T = \frac{N \cdot L}{E \cdot A} \quad \dots\dots\dots (3)$$

: (2) (3)

$$\alpha_p \cdot L \cdot \Delta T = \frac{N \cdot L}{E \cdot A} \quad \dots\dots\dots (4)$$

$$\therefore N = \alpha_p \cdot A \cdot E \cdot \Delta T \quad \dots\dots\dots (5)$$

(5)

(Simply Supported)

[7] (6)

(Orifice Plate)

(Constant Heat Flux)

$$E = [1.09939 \cdot 10^{-2} - 2.917459 \cdot 10^{-2} \cdot (T) - 5.7084838 \cdot 10^{-5} \cdot (T^2) + 3.75691996 \cdot 10^{-9} \cdot (T^3)] \cdot 10^9 \quad \dots\dots\dots (6)$$

(Beam
(Shear

Theory)
Deformation)

(Rotatory Inertia)

[9],[7]

(Conduction

)

$$W(x,t) = \left[\begin{array}{l} -a_1(m_f V_f V_j + PA_f) \left(\frac{n\pi}{L}\right)^2 \sin\left(\frac{n\pi}{L} X\right) - \\ a_2(2m_f V_f) \left(\frac{n\pi}{L}\right) \omega \cos\left(\frac{n\pi}{L} X\right) \\ a_1(2m_f V_f) \left(\frac{n\pi}{L}\right) \omega \cos\left(\frac{n\pi}{L} X\right) - \\ a_2(m_f V_f V_j + PA_f) \left(\frac{n\pi}{L}\right)^2 \sin\left(\frac{n\pi}{L} X\right) \end{array} \right]^2 \Bigg\}^{1/2} \quad (11)$$

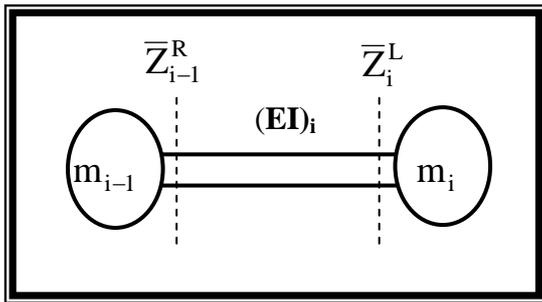
$$V_j = C_v \sqrt{\frac{2(P_1 - P_2)/\rho}{1 - \left(\frac{A_2}{A_p}\right)^2}} \quad \dots\dots(12)$$

(C_v) [11] (0.97-0.99)

(Field Matrix)

(1) (m_{i-1}) (m_i)
(Field Matrix)

$$[\bar{Z}]^L = [\bar{F}] \cdot [\bar{Z}]^R_{i-1} \quad \dots\dots(13)$$



(Element) (1)

(Point Matrix)

(Lumped Mass)

(Convection)

$$q = h(T_w - T_b) \quad \dots\dots(7)$$

(T_i)

(N_{Ud})

: [1]

$$N_{Ud} = \frac{h \cdot d_o}{k_F} = 4.36 \quad \dots(8)$$

$$h = \frac{4.36 \cdot k_F}{d_o} \quad \dots\dots(9)$$

(7)

$$T_{wi} = T_{bi} + \frac{q}{h} \quad \dots\dots(10)$$

(Compressive Force) -1

(Buckling Instability)
(Rotational Force) -2

(1)

$$W(x,t) = (m_f \cdot V_f V_j + PA_f) \frac{\partial^2 Y}{\partial X^2} + (2m_f V_f) \frac{\partial^2 Y}{\partial X \partial t} \quad \dots\dots (11)$$

(Ansys Ver. 9)
(Pipe 16)
(22)

(23)

(Pully or

(Bearing)

Flywheel)

$$[\bar{Z}]_i^R = [\bar{P}] \cdot [\bar{Z}]_i^L$$

.....(14)

(2) , (1)

(Amplitude Vibration)

(20 kW/m²,10 kW/m²)
(250 ≤ Re ≤ 1500)

(Thermal

[13] [12]

Forces)

(12.7mm)

(10.7mm)

(1m)

(5.35mm)

(Flexible – Flexible)

(8733 Kg /mm³)

(Simply Supported)

(E)

(6)

(Simply

-1

Supported)

(Flexible – Flexible)

-2

(3) (2)

[7](11.86*10⁴ N/m²)

(Fully

-3

Developed)

(250 ≤ Re ≤ 1500)

-4

(Heat Flux)

kW/m² (10 ≤ q ≤ 20)

-5

(20)

(21)

(mode shape)

(, , ,)

(y)

(ANSYS 9)

(Ansys . 9) (4-a) (94.24 rad/s)
(q = 0 kW/m²)

(4) (3) (120 (4-b) rad/s)
(q = 0 kW/m²)

(Ansys. 9) (4-a) (94.24rad/s)

(Ansys 9) (F.E.M) (4-c) (126.9 rad/s)

(T.M.M)

(6) (5) (157.07 (130.4 rad/s) rad/s)

-1 (4-e) (4-d)

(Mode Shape) (Flexible-Support)

(Simply-Support)

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q (kW/m ²)	V _f = 0.0306 (m/s)					
	Simply Support (rad/s)			Flexible Support (rad/s)		
	ω ₁	ω ₂	ω ₃	ω ₁	ω ₂	ω ₃
0	126.9	508.9	1137.26	125.66	485.06	1020.38
10	125.66	503.9	1128.46	118.12	459.93	980.18
20	124.4	500.14	1118.41	111.84	441.08	950

(1)

.(V_f = 0.0306 (m/s))

q (kW/m ²)	V _f = 0.122 (m/s)					
	Simply Support (rad/s)			Flexible Support (rad/s)		
	ω ₁	ω ₂	ω ₃	ω ₁	ω ₂	ω ₃
0	126.9	508.9	1137.26	125.66	485.06	1020.38
10	125.66	505.17	1129.71	118.12	461.19	982.69
20	124.4	501.39	1120.92	113.09	443.59	955.04

(2)

.(V_f = 0. 122 (m/s))

q(kW/m ²)		0	10	20
First Mode ω ₁ (rad/s)	T.M.M	126.9	125.6	124.4
	Ansys(FEM)	126.9	125.6	124.4
	Error	0	0	0
Second Mode ω ₂ (rad/s)	T.M.M	508.9	503.9	500.14
	Ansys(FEM)	-----	-----	-----
	Error	-----	-----	-----
Third Mode ω ₃ (rad/s)	T.M.M	1137.26	1128.46	1118.41
	Ansys(FEM)	1137.26	1129.7	1119.66
	Error	0	0.1	0.1

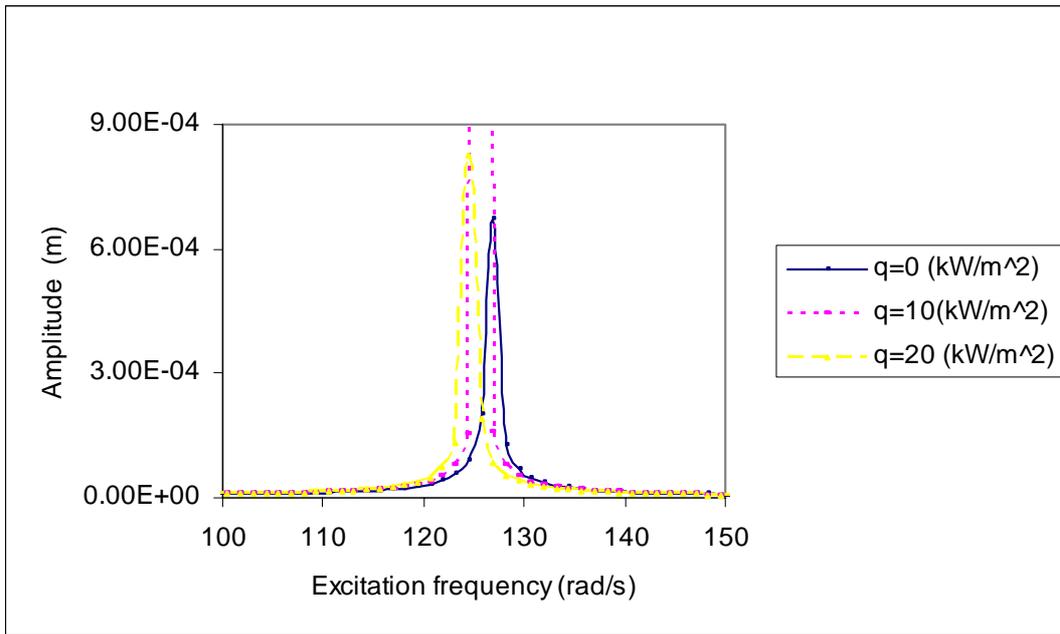
(Ansys 9)

(3)

q (kW/m ²)	First Mode Deflection (m)		Third Mode Deflection (m)	
	T.M.M	Ansys(FEM)	T.M.M	Ansys(FEM)
0	-6.53*10 ⁻⁴	-5.63*10 ⁻³	-6.07*10 ⁻⁵	-3.47*10 ⁻⁵
10	1.05*10 ⁻²	-5.79*10 ⁻⁴	-1.7*10 ⁻⁴	4.1*10 ⁻⁵
20	8*10 ⁻⁴	-3.79*10 ⁻⁴	3.18*10 ⁻⁵	1.58*10 ⁻⁴

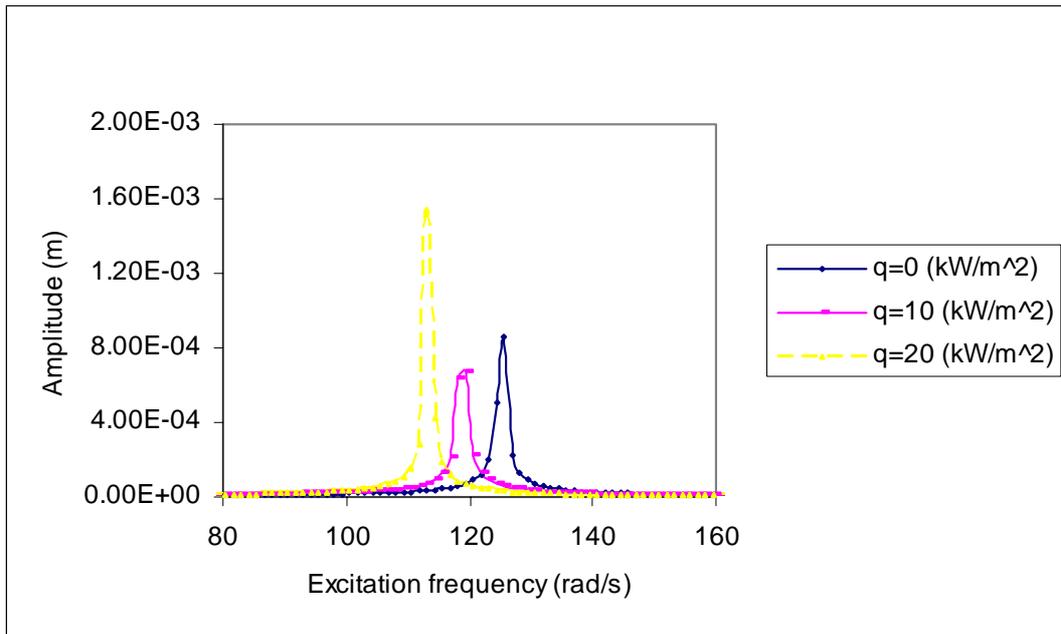
(Ansys 9)

(4)



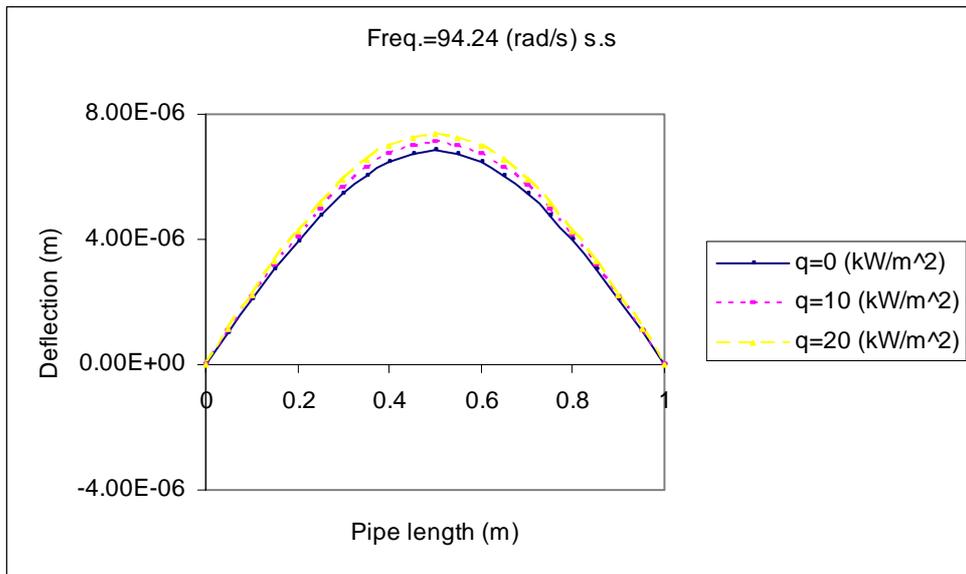
(2)

.(V_f = 0.0306 m/s)

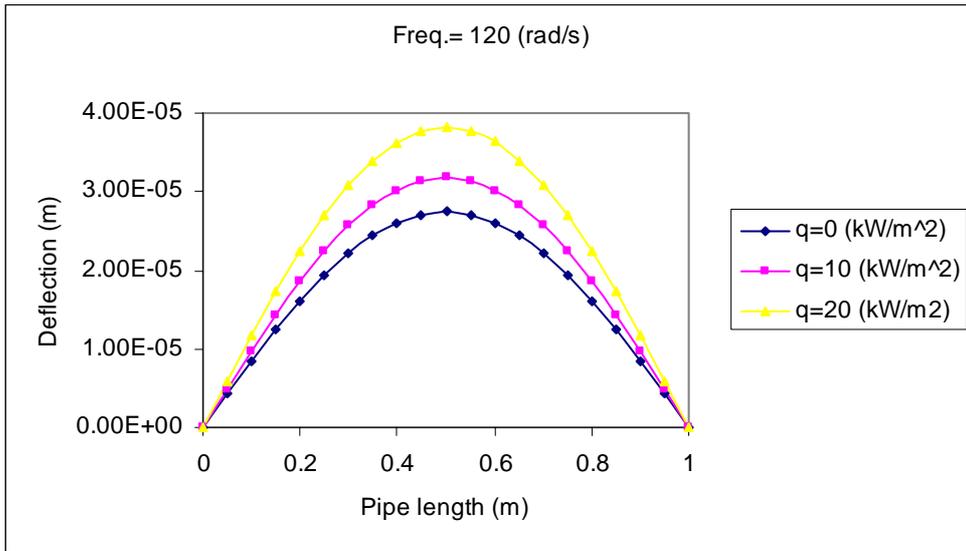


(3)

.(V_f=0.0306 m/s)

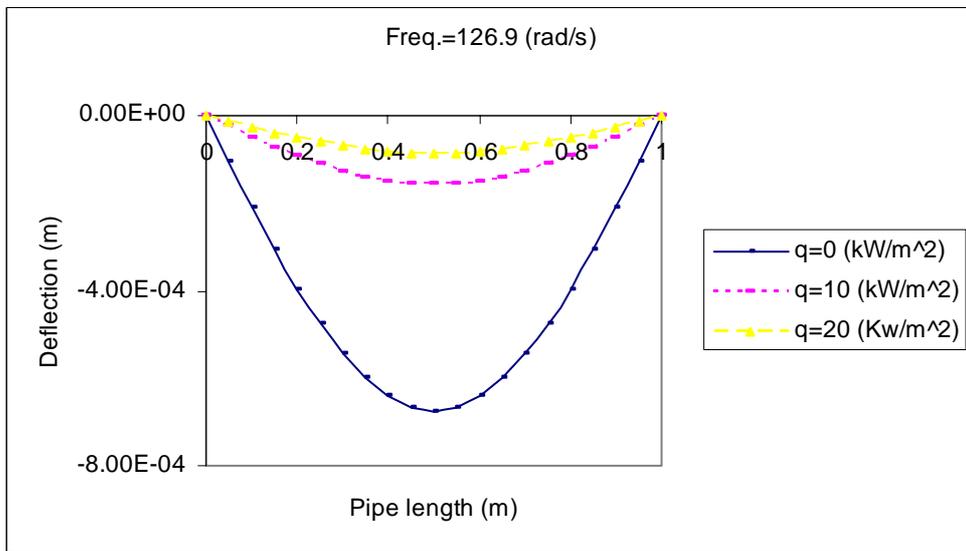


(a)

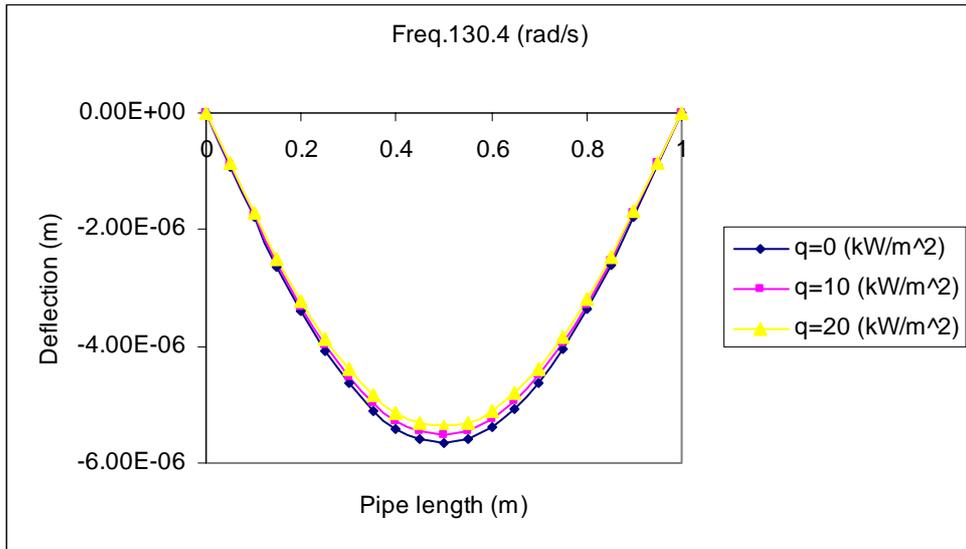


(b)

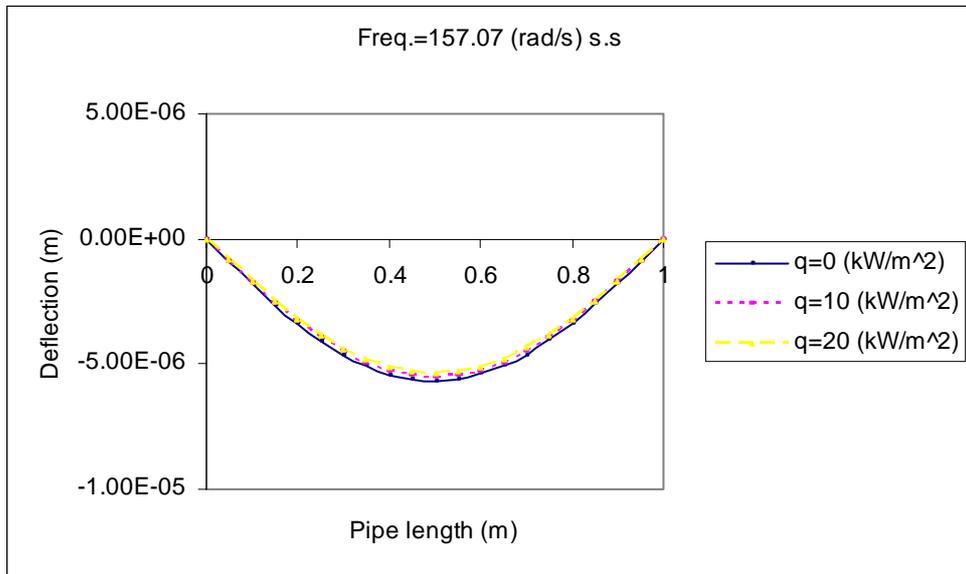
(V_f = 0.0306 m/s) (mode shape) (Simply-Support) (4)



(c)

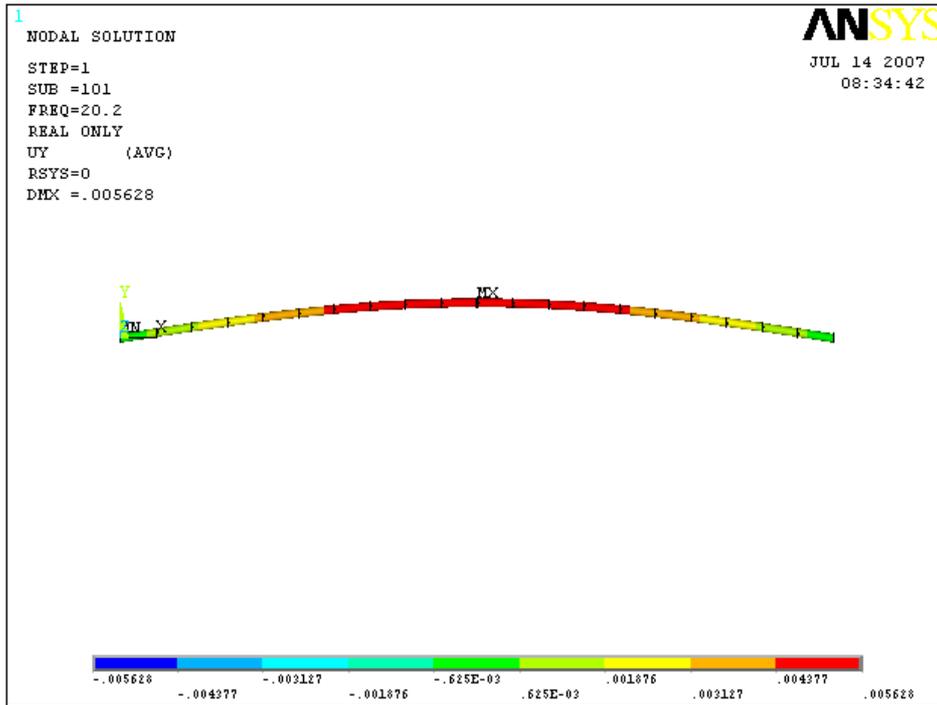


(d)

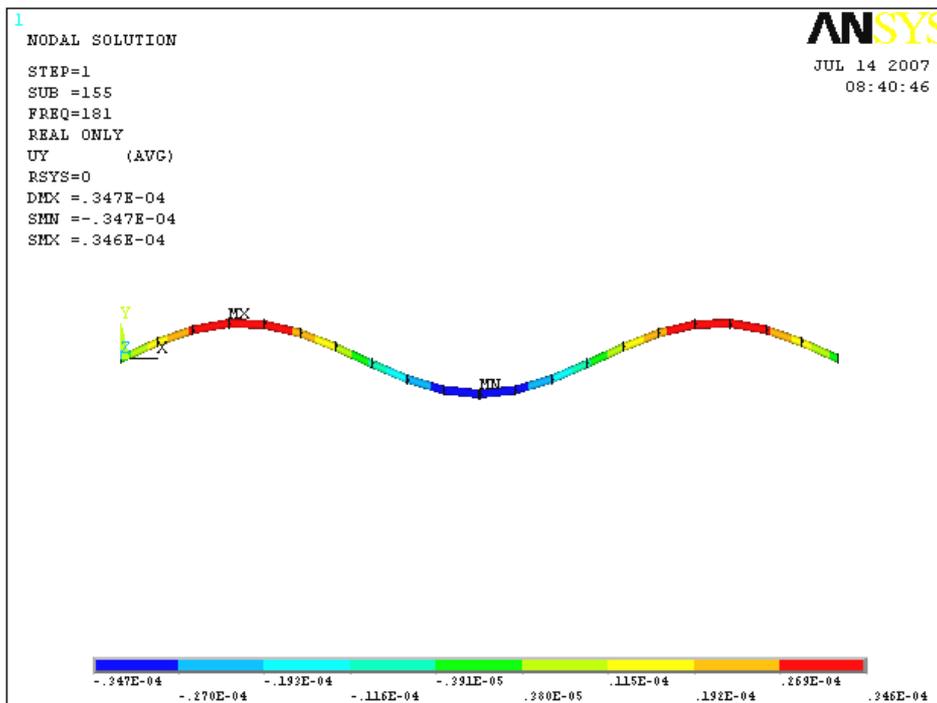


(e)

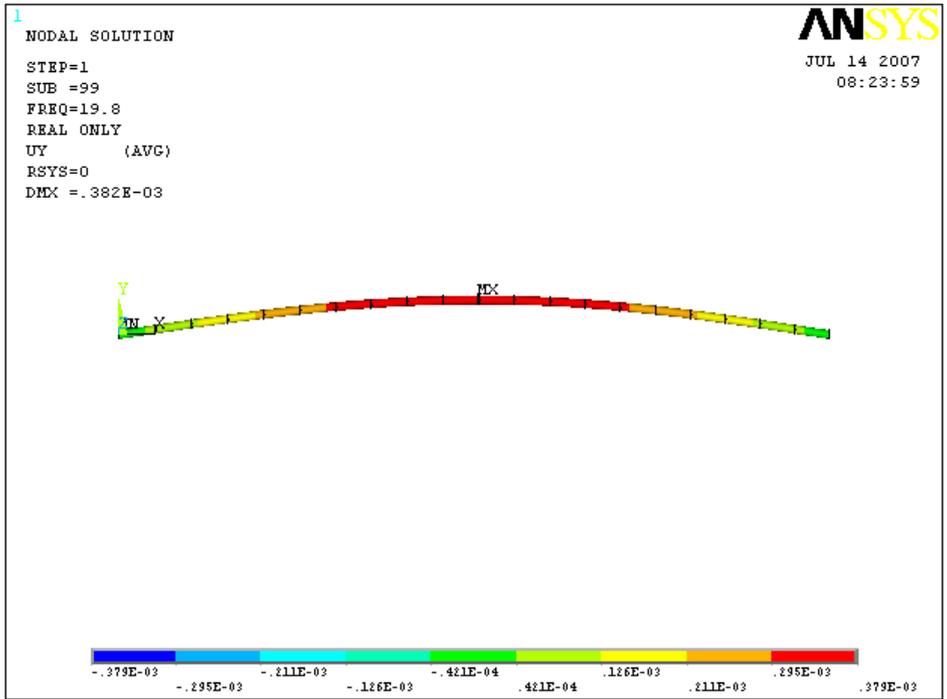
(4)



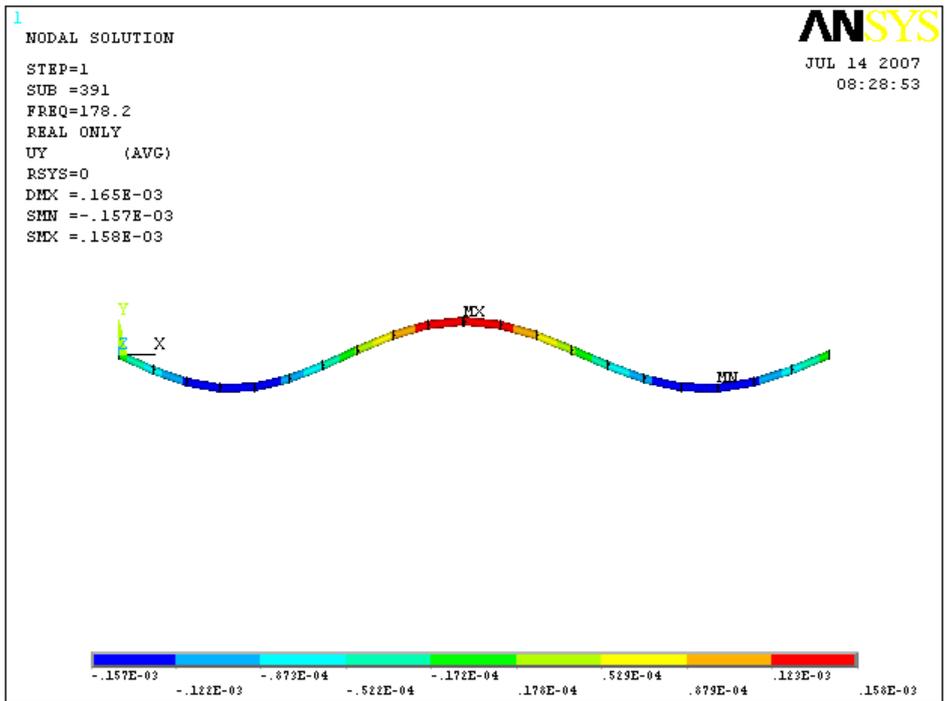
(a)



(b)



(a)



(b)

$q=20 \text{ (kW/m}^2\text{)}$

(6)